

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

2000-03-22 Boeing: Amendment 39-11582. Docket 98-NM-339-AD.

Applicability: Model 747-100, -200, and 747SP series airplanes; line numbers 1 through 567 inclusive; equipped with aluminum diagonal brace underwing fittings; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (f) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent loss of the underwing fitting load path due to missing, damaged, or broken taperlock bolts, which could result in separation of the engine and strut from the airplane, accomplish the following:

Repetitive Inspections

(a) Prior to the accumulation of 9,000 total flight cycles, or within 18 months after the effective date of this AD, whichever occurs later, accomplish the actions required by paragraphs (a)(1) and (a)(2) of this AD in accordance with Boeing Alert Service Bulletin 747-57A2308, dated August 6, 1998. Thereafter, repeat the inspections at intervals not to exceed 18 months until accomplishment of the actions specified in paragraph (d) of this AD.

(1) Perform a detailed visual inspection to detect missing taperlock bolts in the diagonal brace underwing fitting at the Number 1 and Number 4 pylons.

Note 2: For the purposes of this AD, a detailed visual inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally

supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

(2) Perform an ultrasonic inspection to detect damaged or broken taperlock bolts in the diagonal brace underwing fitting at the Number 1 and Number 4 pylons.

Corrective Actions

(b) If any missing, damaged, or broken taperlock bolt is detected during any inspection required by paragraph (a) of this AD, prior to further flight, perform the applicable corrective actions (i.e., inspection, drill/ream, and replacement) in accordance with Boeing Alert Service Bulletin 747-57A2308, dated August 6, 1998; except as provided in paragraph (c) of this AD. Replacement of any taperlock bolt with a new bolt in accordance with this paragraph constitutes terminating action for the repetitive inspections required by paragraph (a) of this AD for that bolt only.

(c) If any crack is detected during the inspection required by paragraph (b) of this AD and the damage to a bolt hole exceeds first oversize (for 0.5-inch bolts) or second oversize (for 0.4375-inch bolts); and the service bulletin specifies to contact Boeing for appropriate action: Prior to further flight, repair in accordance with a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA, Transport Airplane Directorate; or in accordance with a Boeing Company Designated Engineering Representative who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved by the Manager, Seattle ACO, as required by this paragraph, the Manager's approval letter must specifically reference this AD.

Terminating Action

(d) Within 48 months after the effective date of this AD, accomplish the actions required by paragraphs (d)(1) and (d)(2) of this AD in accordance with Boeing Alert Service Bulletin 747-57A2308, dated August 6, 1998. Accomplishment of the actions specified in this paragraph constitutes terminating action for the repetitive inspection requirements of this AD.

(1) Prior to accomplishing the replacement required by paragraph (d)(2) of this AD, perform an open hole high frequency eddy current inspection to detect cracks at the bolt hole locations of the aft 10 taperlock bolts. If any cracking is detected, prior to further flight, perform applicable corrective actions in accordance with paragraph (c) of this AD.

(2) Replace the aft 10 taperlock bolts with new bolts in the diagonal brace underwing fitting at the Number 1 and Number 4 pylons.

Note 3: Accomplishment of the replacement of the diagonal brace underwing fitting in accordance with Figures 5 through 9 of Boeing Service Bulletin 747-57-2288, Revision 1, dated June 26, 1997; or the clearance adjustment in accordance with Figures 10 through 14 of that service bulletin; is acceptable for compliance with the requirements of paragraph (d) of this AD.

Spares

(e) As of the effective date of this AD, no person shall install a bolt, part number BACB30PE() * (), or any other bolt made of 4340, 8740, or PH13-8 Mo steel, in the locations specified in this AD, on any airplane listed in the applicability of this AD.

Alternate Method of Compliance

(f) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle ACO. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

Note 4: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permits

(g) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(h) Except as provided in paragraph (c) of this AD, the actions shall be done in accordance with Boeing Alert Service Bulletin 747-57A2308, dated August 6, 1998. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Boeing Commercial Airplane Group, P. O. Box 3707, Seattle, Washington 98124-2207. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(i) This amendment becomes effective on March 28, 2000.

Issued in Renton, Washington, on February 11, 2000.

Donald L. Riggins,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 00-3797 Filed 2-18-00; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 95-NM-150-AD; Amendment 39-11580; AD 2000-03-20]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A300-600 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to all Airbus Model A300–600 series airplanes, that requires repetitive ultrasonic inspections to detect cracks on the forward fittings in the radius of frame 40 adjacent to the tension bolts in the center section of the wings, and various follow-on actions. This amendment is prompted by reports of cracking due to fatigue-related stress in the radius of frame 40 adjacent to the tension bolts at the center/outer wing junction. The actions specified by this AD are intended to detect and correct fatigue cracking on the forward fittings in the radius of frame 40 adjacent to the tension bolts in the center section of the wings, which could result in reduced structural integrity of the wings.

DATES: Effective March 28, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of March 28, 2000.

ADDRESSES: The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Norman B. Martenson, Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055–4056; telephone (425) 227–2110; fax (425) 227–1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all Airbus Model A300–600 series airplanes was published as a supplemental notice of proposed rulemaking (NPRM) in the *Federal Register* on November 24, 1998 (63 FR 64918). That action proposed to require repetitive ultrasonic inspections to detect cracks on the forward fittings in the radius of frame 40 adjacent to the tension bolts in the center section of the wings, and various follow-on actions.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due

consideration has been given to the comments received.

One commenter, an operator, has no comments on the proposed supplemental NPRM, except to report that the ultrasonic inspections have been accomplished on part of its fleet, with no findings of cracking as yet.

Clarification of Flight Hour Thresholds

One commenter, the manufacturer, states that it cannot determine how the flight hour thresholds specified in the supplemental NPRM were calculated by the FAA. The FAA infers that the commenter is requesting clarification regarding the requirement to accomplish the initial inspection at “7,250 total landings or 17,700 total flight hours, whichever occurs first,” as specified in paragraph (a)(1)(i) of the supplemental NPRM.

The FAA herewith provides the requested information. The threshold of 7,600 total landings specified in Airbus Service Bulletin A300–57–6062, Revision 02, dated January 29, 1997, is calculated using the “adjustment-for-range” formula and a fatigue rating of 0.13, and is valid for airplanes having an average flight time (AFT) of 2.1 hours. (Airbus Service Bulletin A300–57–6062, Revision 02, was referenced in the supplemental NPRM as the appropriate source of service information for this AD.) At the time the supplemental NPRM was developed, the AFT for all Airbus Model A300–600 airplanes operated in the United States was 2.45 hours. The FAA used the 2.45 AFT and the “adjustment-for-range” formula to obtain an adjustment factor appropriate for airplanes on the U.S. Register, as follows:

$$1 - [(2.45 - 2.1) * 0.13] = 0.9545$$
$$0.9545 * 7,600 = 7,254.$$

Additionally, the FAA multiplied the 2.45 AFT by the adjusted landing threshold to obtain an approximation of the corresponding flight hours, as follows:

$$7,254 * 2.45 = 17,773.$$

Using these calculations, the FAA determined appropriate landing and flight-hour thresholds, previously described, for the requirements of this AD.

Request To Revise Compliance Threshold

The same commenter requests that the supplemental NPRM be revised to replace the compliance thresholds with thresholds calculated using a new method. The commenter states that the flight-hour thresholds specified in the supplemental NPRM seem to be overly conservative for operators utilizing

airplanes in long-range operations; such thresholds will penalize those operators by requiring inspections earlier than necessary. The commenter states that, in order to avoid using the current “adjustment-for-range” formula for calculating compliance times, an alternative method has been developed. This new method will provide constant flight-hour thresholds and intervals in addition to the flight cycle-based thresholds and intervals. The commenter proposes to provide these additional thresholds to the FAA, calculated in accordance with the new method.

The FAA does not concur. The FAA has reviewed the alternative method proposed by the manufacturer for calculating flight-hour thresholds, and has discussed the requirements of this AD further with the manufacturer and with the Direction Generale de l'Aviation Civile (DGAC), which is the airworthiness authority for France. The FAA concurs technically with use of such flight-hour and flight-cycle thresholds and intervals. However, the revised thresholds have not yet been made available to the FAA for its review, and the manufacturer advises that there has been a delay in internal review and approval of the thresholds. The FAA does not consider it appropriate to further delay issuance of this AD while awaiting the receipt of these data. Therefore, the FAA has determined that it is necessary to issue the AD with the currently specified thresholds. Once revised thresholds are made available, the FAA will review them and determine if further rulemaking is necessary. Additionally, under the provisions of paragraph (d)(1) of the AD, the FAA may approve requests for adjustments to the compliance time if data are submitted to substantiate that such adjustments of the compliance time would provide an acceptable level of safety.

Conclusion

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

The FAA estimates that 35 airplanes of U.S. registry will be affected by this AD. It will take approximately 2 work hours per airplane (1 work hour per side) to accomplish the required ultrasonic inspection, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be

\$4,200, or \$120 per airplane, per inspection cycle.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

2000-03-20 Airbus Industrie: Amendment 39-11580. Docket 95-NM-150-AD.

Applicability: All Model A300-600 airplanes, certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability

provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d)(1) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect and correct fatigue cracking on the forward fittings in the radius of frame 40 adjacent to the tension bolts in the center section of the wings, which could result in reduced structural integrity of the wings, accomplish the following:

Inspections and Corrective Actions

(a) Perform an ultrasonic inspection to detect cracking on the forward fittings in the radius of frame 40 adjacent to the tension bolts in the center section of the wings, in accordance with Airbus Service Bulletin A300-57-6062, Revision 02, dated January 29, 1997, at the applicable time specified in either paragraph (a)(1) or (a)(2) of this AD.

(1) For airplanes that have accumulated fewer than 9,100 total landings or 22,300 total flight hours as of the effective date of this AD: Inspect at the later of the times specified in either paragraph (a)(1)(i) or (a)(1)(ii) of this AD.

(i) Prior to the accumulation of 7,250 total landings or 17,700 total flight hours, whichever occurs first.

(ii) Within 1,500 landings after the effective date of this AD.

(2) For airplanes that have accumulated 9,100 total landings or more and 22,300 total flight hours or more as of the effective date of this AD: Inspect within 750 landings after the effective date of this AD.

Note 2: Inspections that were accomplished prior to the effective date of this AD in accordance with Airbus Service Bulletin A300-57-6062, Revision 1, dated July 23, 1995, are considered acceptable for compliance with paragraph (a) of this AD.

(b) If no crack is detected during the inspection required by paragraph (a) of this AD, repeat the ultrasonic inspection required by that paragraph thereafter at intervals not to exceed 6,500 landings or 16,000 flight hours, whichever occurs first; in accordance with Airbus Service Bulletin A300-57-6062, Revision 02, dated January 29, 1997.

(c) If any crack is detected during any inspection required by paragraph (a) or (b) of this AD, prior to further flight, install an access door, and perform an eddy current inspection to confirm the presence of a crack; in accordance with Airbus Service Bulletin A300-57-6062, Revision 02, dated January 29, 1997. Accomplishment of this eddy current inspection terminates the repetitive inspection requirement of paragraph (b) of this AD.

(1) If no crack is detected during the eddy current inspection, repeat the eddy current

inspection, in accordance with the service bulletin, thereafter at intervals not to exceed 6,500 landings or 16,000 flight hours, whichever occurs first.

(2) If any crack is detected during any eddy current inspection performed in accordance with paragraph (c) or (c)(1) of this AD, prior to further flight, blend out the crack and repeat the eddy current inspection in accordance with the service bulletin.

(i) If the eddy current inspection performed after the blend-out shows that the crack has been removed, and if the blend-out is equal to or less than 50 millimeters (mm) long and equal to or less than 2 mm deep, thereafter repeat the eddy current inspection at intervals not to exceed 2,800 landings or 7,000 flight hours, whichever occurs first.

(ii) If the eddy current inspection performed after the blend-out shows that the crack has not been removed, or if the blend-out is more than 50 mm long or more than 2 mm deep, prior to further flight, repair in accordance with a method approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate; or the Direction Generale de l'Aviation Civile (or its delegated agent).

Alternative Methods of Compliance

(d)(1) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

(2) Operators may request an extension to the compliance times of this AD in accordance with the "adjustment-for-range" formula found in Paragraph 1.B.(5) of Airbus Service Bulletin A300-57-6062, Revision 02, dated January 29, 1997; and provided in A300-600 Maintenance Review Board, Section 5, Paragraph 5.4. The average flight time per flight cycle (landing) in hours used in this formula should be for an individual airplane. Average flight time for a group of airplanes may be used if all airplanes of the group have flight times differing by no more than 10 percent. If compliance times are based on the average flight time for a group of airplanes, the flight times for individual airplanes of the group must be included for FAA review.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

Special Flight Permits

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(f) Except as required by (c)(2)(ii) of this AD, the actions shall be done in accordance with Airbus Service Bulletin A300-57-6062,

Revision 02, dated January 29, 1997, which contains the specified effective pages:

Page No.	Revision level shown on page	Date shown on page
1–34, 37	02	Jan. 29, 1999
35, 36	1	Jul. 23, 1995

This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 4: The subject of this AD is addressed in French airworthiness directive 95–063–177(B)R3, dated July 2, 1997.

(g) This amendment becomes effective on March 28, 2000.

Issued in Renton, Washington, on February 11, 2000.

Donald L. Riggins,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 00–3796 Filed 2–18–00; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 98–NM–193–AD; Amendment 39–11581; AD 2000–03–21]

RIN 2120–AA64

Airworthiness Directives; Boeing Model 767 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Boeing Model 767 series airplanes, that requires a one-time detailed visual inspection to detect discrepancies of the wire expando sleeve of the wire bundles adjacent to the landing gear control lever module; certain follow-on actions and repair, if necessary; and wrapping the wire expando sleeve with tape, or with zippertubing and tape. This amendment is prompted by reports indicating that the landing gear failed to extend on an in-service airplane, and that the landing gear control cable was severed on a second in-service airplane. The actions specified by this AD are intended to

prevent interference between the landing gear control lever and wire bundles adjacent to the landing gear control lever module, and to prevent wire chafing and arcing between the landing gear control cable and adjacent wire bundles, which could result in the inability to extend the landing gear prior to landing.

DATES: Effective March 28, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of March 28, 2000.

ADDRESSES: The service information referenced in this AD may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124–2207. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Elias Natsiopoulou, Aerospace Engineer, Systems and Equipment Branch, ANM–130S, FAA, Transport Airplane Directorate, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055–4056; telephone (425) 227–1279; fax (425) 227–1181.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Boeing Model 767 series airplanes was published in the **Federal Register** on February 17, 1999 (64 FR 7829). That action proposed to require a one-time inspection to detect discrepancies of the wire expando sleeve of the wire bundles adjacent to the landing gear control lever module; certain follow-on actions and repair, if necessary; and wrapping the wire expando sleeve with tape, or with zippertubing and tape.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received.

Support for the Proposal

Two commenters concur with the proposal. Another commenter states that it is not affected by the proposal, as the proposed actions have been accomplished for its fleet.

Request to Clarify Wire Bundle Interference

One commenter, the manufacturer, recommends clarification of the description of wire bundle interference in the Summary and Discussion sections of the proposed rule. The notice of proposed rulemaking (NPRM) describes two incidents on in-service Model 767 series airplanes. In the first incident, the expando sleeve on a wire bundle adjacent to the landing gear lever mechanism in the flight deck became caught on the lever mechanism such that the lever could not be moved from the UP position, and the gear was extended by depressurizing the center hydraulic system. The commenter contends that interference of the wire bundle expando sleeve with the landing gear control lever did not result in the wires for the alternate extension system interfering and arcing with the landing gear UP cable. Such interference and arcing are unrelated events. The contact with the gear UP cable was solely the result of the amount of slack in the wire bundle itself. The only wire bundle long enough to reach the gear UP cable is the one that contains wires for the alternate extension system.

In the second incident, which was an unrelated incident, a wire bundle containing wires for the alternate extension system chafed on the landing gear UP cable, causing arcing and failure of the gear UP cable; however, the landing gear was extended at the time and gear extension capability was still available through the gear DOWN cable. The commenter contends that the inability to extend the landing gear also is not an issue if the landing gear had been UP and locked; the landing gear will extend. If the landing gear had been retracted when the UP cable was severed, and assuming that the alternate extension system had been rendered inoperative due to the arcing, the normal extension system is available. When gear DOWN is selected, the landing gear selector valve will move to