

# Rules and Regulations

Federal Register

Vol. 80, No. 206

Monday, October 26, 2015

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2015-4209; Directorate Identifier 2015-NM-156-AD; Amendment 39-18302; AD 2015-21-09]

RIN 2120-AA64

#### Airworthiness Directives; The Boeing Company Airplanes

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule; request for comments.

**SUMMARY:** We are superseding Airworthiness Directive (AD) 2015-19-02 for all The Boeing Company Model 767 airplanes. AD 2015-19-02 required revising the maintenance or inspection program to include new airworthiness limitations. This AD continues to require a maintenance or inspection program revision, but with revised language. This AD was prompted by a determination that certain language in the airworthiness limitation was not accurate in AD 2015-19-02. We are issuing this AD to detect and correct latent failures of the fuel shutoff valve to the engine and auxiliary power unit (APU), which could result in the inability to shut off fuel to the engine and APU and, in case of certain fires, an uncontrollable fire that could lead to structural failure.

**DATES:** This AD is effective October 28, 2015.

We must receive any comments on this AD by December 10, 2015.

**ADDRESSES:** You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-4209; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:** Rebel Nichols, Aerospace Engineer, Propulsion Branch, ANM-140S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue SW., Renton, WA 98057-3356; phone: 425-917-6509; fax: 425-917-6590; email: [rebel.nichols@faa.gov](mailto:rebel.nichols@faa.gov).

#### SUPPLEMENTARY INFORMATION:

##### Discussion

On September 7, 2015, we issued AD 2015-19-02, Amendment 39-18265 (80 FR 55512, September 16, 2015), for all The Boeing Company Model 767 airplanes. AD 2015-19-02 required revising the maintenance or inspection program to include new airworthiness limitations. AD 2015-19-02 resulted from reports of latently failed fuel shutoff valves discovered during fuel filter replacement. We issued AD 2015-19-02 to detect and correct latent failures of the fuel shutoff valve to the engine and APU, which could result in the inability to shut off fuel to the engine and APU and, in case of certain fires, an uncontrollable fire that could lead to structural failure.

#### Actions Since AD 2015-19-02, Amendment 39-18265 (80 FR 55512, September 16, 2015), Was Issued

Since we issued AD 2015-19-02, Amendment 39-18265 (80 FR 55512, September 16, 2015), we have determined that extraneous language was included in two locations of the text of the airworthiness limitations specified in AD 2015-19-02. In paragraph C.7.a. of the "Description" column of figure 1 to paragraph (g) of AD 2015-19-02, the text "or the APU selector switch on the overhead panel is in the ON position" is not relevant to the actions specified in that paragraph. In paragraph A.5. of the "Description" column of figure 3 to paragraph (g) of AD 2015-19-02, the text "the FUEL CONTROL switch is in the RUN position or" is not relevant to the actions specified in that paragraph.

We have determined that the language must be corrected to avoid any confusion in the paragraphs of the airworthiness limitation. We are issuing this AD to detect and correct latent failures of the fuel shutoff valve to the engine and APU, which could result in the inability to shut off fuel to the engine and APU and, in case of certain fires, an uncontrollable fire that could lead to structural failure.

#### FAA's Determination

We are issuing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or develop in other products of the same type design.

#### AD Requirements

This AD requires revising the maintenance or inspection program to include new airworthiness limitations.

#### Interim Action

We consider this AD interim action. The manufacturer is currently developing a modification that will address the unsafe condition identified in this AD. Once this modification is developed, approved, and available, we might consider additional rulemaking.

#### FAA's Justification and Determination of the Effective Date

We are superseding AD 2015-19-02, Amendment 39-18265 (80 FR 55512, September 16, 2015), to correct inaccurate terminology in the

“Description” column of figure 1 to paragraph (g) of AD 2015–19–02 and figure 3 to paragraph (g) of AD 2015–19–02. We have made no other changes to the requirements published in AD 2015–19–02. We have determined that the changes impose no additional burden on any operator. Therefore, we find that notice and opportunity for prior public comment are unnecessary and that good cause exists for making this amendment effective in less than 30 days.

Comments Invited

This AD is a final rule that involves requirements affecting flight safety, and

we did not provide you with notice and an opportunity to provide your comments before it becomes effective. However, we invite you to send any written data, views, or arguments about this AD. Send your comments to an address listed under the ADDRESSES section. Include the Docket Number FAA–2015–4209 and Directorate Identifier 2015–NM–156–AD at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this AD. We will consider all comments received by the closing date and may

amend this AD because of those comments. We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this AD.

Costs of Compliance

We estimate that this AD affects 450 airplanes of U.S. registry. We estimate the following costs to comply with this AD:

ESTIMATED COSTS

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Incorporating Airworthiness Limitation .....	1 work-hour × \$85 per hour = \$85 .....	\$0	\$85	\$38,250

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, “General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

(1) Is not a “significant regulatory action” under Executive Order 12866,

(2) Is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),

- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:  
Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD) 2015–19–02, Amendment 39–18265 (80 FR 55512, September 16, 2015), and adding the following new AD:  
2015–21–09 The Boeing Company: Amendment 39–18302; Docket No. FAA–2015–4209; Directorate Identifier 2015–NM–156–AD.

(a) Effective Date

This AD is effective October 28, 2015.

(b) Affected ADs

This AD replaces AD 2015–19–02, Amendment 39–18265 (80 FR 55512, September 16, 2015).

(c) Applicability

This AD applies to all The Boeing Company Model 767–200, –300, –300F, and –400ER series airplanes, certificated in any category.

(d) Subject

Air Transport Association (ATA) of America Code 28, Fuel.

(e) Unsafe Condition

This AD was prompted by reports of latently failed fuel shutoff valves discovered during fuel filter replacement. We are issuing this AD to detect and correct latent failures of the fuel shutoff valve to the engine and auxiliary power unit (APU), which could result in the inability to shut off fuel to the engine and APU and, in case of certain fires, an uncontrollable fire that could lead to structural failure.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Revision of Maintenance or Inspection Program

Within 30 days after the effective date of this AD, revise the maintenance or inspection program, as applicable, to add airworthiness limitation numbers 28–AWL–ENG, 28–AWL–MOV, and 28–AWL–APU, by incorporating the information specified in figure 1, figure 2, and figure 3 to paragraph (g) of this AD into the Airworthiness Limitations Section of the Instructions for Continued Airworthiness. The initial compliance time for accomplishing the actions specified in figure 1, figure 2, and figure 3 to paragraph (g) of this AD is within 10 days after accomplishing the maintenance or inspection program revision required by this paragraph.

FIGURE 1 TO PARAGRAPH (g) OF THIS AD: ENGINE FUEL SHUTOFF VALVE (FUEL SPAR VALVE) POSITION INDICATION OPERATIONAL CHECK

AWL No.	Task	Interval	Applicability	Description
28-AWL-ENG ...	ALI	DAILY .....  INTERVAL NOTE: The operational check is not required on days when the airplane is not used in revenue service. The check must be done before further flight once the airplane is returned to revenue service.	767–200, –300, and –300F airplanes.  APPLICABILITY NOTE: Applies to airplanes with an actuator installed at the engine fuel spar valve position having part number (P/N) MA20A2027 (S343T003–56) or P/N MA30A1001 (S343T003–66).	Engine Fuel Shutoff Valve (Fuel Spar Valve) Position Indication Operational Check.  Concern: The fuel spar valve actuator design can result in airplanes operating with a failed fuel spar valve actuator that is not reported. A latently failed fuel spar valve actuator could prevent fuel shutoff to an engine. In the event of certain engine fires, the potential exists for an engine fire to be uncontrollable.  Perform one of the following checks/inspection of the fuel spar valve position (unless checked by the flightcrew in a manner approved by the principal operations inspector): A. Operational Check during engine shutdown. 1. Do an operational check of the left engine fuel spar valve actuator. a. As the L FUEL CONTROL switch on the quadrant control stand is moved to the CUTOFF position, verify the left SPAR VALVE disagreement light on the quadrant control stand illuminates and then goes off. b. If the test fails (light fails to illuminate), before further flight, repair faults as required (refer to Boeing airplane maintenance manual (AMM) 28–22–11). 2. Do an operational check of the right engine fuel spar valve actuator. a. As the R FUEL CONTROL switch on the quadrant control stand is moved to the CUTOFF position, verify the right SPAR VALVE disagreement light on the quadrant control stand illuminates and then goes off. b. If the test fails (light fails to illuminate), before further flight, repair faults as required (refer to Boeing AMM 28–22–11). B. Operational check during engine start. 1. Do an operational check of the left engine fuel spar valve actuator. a. As the L FUEL CONTROL switch on the quadrant control stand is moved to the RUN (or RICH) position, verify the left SPAR VALVE disagreement light on the quadrant control stand illuminates and then goes off. b. If the test fails (light fails to illuminate), before further flight, repair faults as required (refer to Boeing AMM 28–22–11). 2. Do an operational check of the right engine fuel spar valve actuator. a. As the R FUEL CONTROL switch on the quadrant control stand is moved to the RUN (or RICH) position, verify the right SPAR VALVE disagreement light on the quadrant control stand illuminates and then goes off. b. If the test fails (light fails to illuminate), before further flight, repair faults as required (refer to Boeing AMM 28–22–11). C. Operational check without engine operation. 1. Supply electrical power to the airplane using standard practices. 2. Make sure all fuel pump switches on the Overhead Panel are in the OFF position. 3. If the auxiliary power unit (APU) is running, open and collar the L FWD FUEL BOOST PUMP (C00372) circuit breaker on the Main Power Distribution Panel. 4. Make sure LEFT and RIGHT ENG FIRE switches on the Aft Aisle Stand are in the NORMAL (IN) position. 5. Make sure L and R ENG START Selector Switches on the Overhead Panel, are in the OFF position. 6. Do an operational check of the left engine fuel spar valve actuator.

FIGURE 1 TO PARAGRAPH (g) OF THIS AD: ENGINE FUEL SHUTOFF VALVE (FUEL SPAR VALVE) POSITION INDICATION OPERATIONAL CHECK—Continued

AWL No.	Task	Interval	Applicability	Description
				<p>a. Move L FUEL CONTROL switch on the quadrant control stand to the RUN position and wait approximately 10 seconds.</p> <p>NOTE: It is normal under this test condition for the ENG VALVE disagreement light on the quadrant control stand to stay illuminated.</p> <p>b. Move L FUEL CONTROL switch on the quadrant control stand to the CUTOFF position.</p> <p>c. Verify the left SPAR VALVE disagreement light on the quadrant control stand illuminates and then goes off.</p> <p>d. If the test fails (light fails to illuminate), before further flight, repair faults as required (refer to Boeing AMM 28-22-11).</p> <p>7. Do an operational check of the right engine fuel spar valve actuator.</p> <p>a. Move R FUEL CONTROL switch on the quadrant control stand to the RUN position and wait approximately 10 seconds once the FUEL CONTROL switch is in the RUN position.</p> <p>NOTE: It is normal under this test condition for the ENG VALVE disagreement light on the quadrant control stand to stay illuminated.</p> <p>b. Move R FUEL CONTROL switch on the quadrant control stand to the CUTOFF position.</p> <p>c. Verify the right SPAR VALVE disagreement light on the quadrant control stand illuminates and then goes off.</p> <p>d. If the test fails (light fails to illuminate), before further flight, repair faults as required (refer to Boeing AMM 28-22-11).</p> <p>8. If the L FWD FUEL BOOST PUMP circuit breaker was collared in step 3, remove collar and close.</p> <p>D. Perform an inspection of the fuel spar valve actuator position.</p> <p>NOTE: This inspection may be most useful whenever the SPAR VALVE light does not function properly.</p> <p>1. Make sure the L FUEL CONTROL switch on the quadrant control stand is in the CUTOFF position.</p> <p>NOTE: It is not necessary to cycle the FUEL CONTROL switch to do this inspection.</p> <p>2. Inspect the left engine fuel spar valve actuator located in the left rear spar.</p> <p>NOTE: The Fuel Spar Valve actuators are located behind main gear doors on the rear spar.</p> <p>a. Verify the manual override handle on the engine fuel spar valve actuator is in the CLOSED position.</p> <p>b. Repair or replace any fuel spar valve actuator that is not in the CLOSED position (refer to Boeing AMM 28-22-11).</p> <p>3. Make sure the R FUEL CONTROL switch on the quadrant control stand is in the CUTOFF position.</p> <p>NOTE: It is not necessary to cycle the FUEL CONTROL switch to do this inspection.</p> <p>4. Inspect the right engine fuel spar valve actuator located in the right rear spar.</p> <p>NOTE: The Fuel Spar Valve actuators are located behind main gear doors on the rear spar.</p> <p>a. Verify the manual override handle on the engine fuel spar valve actuator is in the CLOSED position.</p> <p>b. Repair or replace any fuel spar valve actuator that is not in the CLOSED position (refer to Boeing AMM 28-22-11).</p>

FIGURE 2 TO PARAGRAPH (G) OF THIS AD: ENGINE FUEL SHUTOFF VALVE (FUEL SPAR VALVE) ACTUATOR INSPECTION

AWL No.	Task	Interval	Applicability	Description
28-AWL-MOV ....	ALI	10 DAYS .....  INTERVAL NOTE: The inspection is not required on days when the airplane is not used in revenue service. The inspection must be done before further flight if it has been 10 or more calendar days since last inspection.	767–400ER series airplanes. APPLICABILITY NOTE: Applies to airplanes with an actuator installed at the engine fuel spar valve position having part number (P/N) MA20A2027 (S343T003–56) or P/N MA30A1001 (S343T003–66).	Engine Fuel Shutoff Valve (Fuel Spar Valve) Actuator Inspection. Concern: The fuel spar valve actuator design can result in airplanes operating with a failed fuel spar valve actuator that is not reported. A latently failed fuel spar valve actuator would prevent fuel shutoff to an engine. In the event of certain engine fires, the potential exists for an engine fire to be uncontrollable. Perform an inspection of the fuel spar valve actuator position. NOTE: The fuel spar valve actuators are located behind main gear doors on the rear spar. 1. Make sure the L FUEL CONTROL switch on the quadrant control stand is in the CUTOFF position. NOTE: It is not necessary to cycle the FUEL CONTROL switch to do this inspection. 2. Inspect the left engine fuel spar valve actuator located in the left rear spar. a. Verify the manual override handle on the engine fuel spar valve actuator is in the CLOSED position. b. Repair or replace any fuel spar valve actuator that is not in the CLOSED position (refer to Boeing AMM 28–22–11). 3. Make sure the R FUEL CONTROL switch on the quadrant control stand is in the CUTOFF position. NOTE: It is not necessary to cycle the FUEL CONTROL switch to do this inspection. 4. Inspect the right engine fuel spar valve actuator located in the right rear spar. a. Verify the manual override handle on the engine fuel spar valve actuator is in the CLOSED position. b. Repair or replace any fuel spar valve actuator that is not in the CLOSED position (refer to Boeing AMM 28–22–11).

FIGURE 3 TO PARAGRAPH (G) OF THIS AD: AUXILIARY POWER UNIT (APU) FUEL SHUTOFF VALVE POSITION INDICATION OPERATIONAL CHECK

AWL No.	Task	Interval	Applicability	Description
28-AWL-APU ...	ALI	10 DAYS .....  INTERVAL NOTE: The operational check is not required on days when the airplane is not used in revenue service. The operational check must be done before further flight with an operational APU if it has been 10 or more calendar days since last check.	ALL .....  APPLICABILITY NOTE: Applies to airplanes with an actuator installed at the APU fuel shutoff valve position having part number (P/N) MA20A2027 (S343T003–56) or MA30A1001 (S343T003–66).	APU Fuel Shutoff Valve Position Indication Operational Check. Concern: The APU fuel shutoff valve actuator design can result in airplanes operating with a failed APU fuel shutoff valve actuator that is not reported. A latently failed APU fuel shutoff valve actuator could prevent fuel shutoff to the APU. In the event of certain APU fires, the potential exists for an APU fire to be uncontrollable. Perform the operational check of the APU fuel shutoff valve position indication (unless checked by the flightcrew in a manner approved by the principal operations inspector). A. Do an operational check of the APU fuel shutoff valve position indication. 1. If the APU is running, unload and shut down the APU using standard practices. 2. Supply electrical power to the airplane using standard practices. 3. Make sure the APU FIRE switch on the Aft Aisle Stand is in the NORMAL (IN) position. 4. Make sure there is at least 1,000 lbs (500 kgs) of fuel in the Left Main Tank. 5. Move APU Selector switch on the Overhead Panel to the ON position and wait approximately 10 seconds once the APU selector switch on the overhead panel is in the ON position. 6. Move the APU Selector switch on the Overhead Panel to the OFF position.

FIGURE 3 TO PARAGRAPH (G) OF THIS AD: AUXILIARY POWER UNIT (APU) FUEL SHUTOFF VALVE POSITION INDICATION OPERATIONAL CHECK—Continued

AWL No.	Task	Interval	Applicability	Description
				<p>7. Verify the APU FAULT light on the Overhead Panel illuminates and then goes off.</p> <p>8. If the test fails (light fails to illuminate), before further flight requiring APU availability, repair faults as required (refer to Boeing AMM 28–25–02).</p> <p>NOTE: Dispatch may be permitted per MMEL 28–25–02 if APU is not required for flight.</p>

**(h) No Alternative Actions or Intervals**

After accomplishment of the maintenance or inspection program revision required by paragraph (g) of this AD, no alternative actions (e.g., inspections) or intervals may be used unless the actions or intervals are approved as an alternative method of compliance (AMOC) in accordance with the procedures specified in paragraph (i)(1) of this AD.

**(i) Alternative Methods of Compliance (AMOCs)**

(1) The Manager, Seattle Aircraft Certification Office (ACO) FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (j) of this AD. Information may be emailed to: *9-ANM-Seattle-ACO-AMOC-Requests@faa.gov*.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

**(j) Related Information**

For more information about this AD, contact Rebel Nichols, Aerospace Engineer, Propulsion Branch, ANM–140S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue SW., Renton, WA 98057–3356; phone: 425–917–6509; fax: 425–917–6590; email: *rebel.nichols@faa.gov*.

**(k) Material Incorporated by Reference**

None.

Issued in Renton, Washington, on October 16, 2015.

**Jeffrey E. Duven,**

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2015–26983 Filed 10–23–15; 8:45 am]

**BILLING CODE 4910–13–P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA–2015–0933; Directorate Identifier 2014–NM–098–AD; Amendment 39–18297; AD 2015–21–05]

RIN 2120–AA64

**Airworthiness Directives; Fokker Services B.V. Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for all Fokker Services B.V. Model F.27 Mark 200, 300, 400, 500, 600, and 700 airplanes. This AD was prompted by a design review, which revealed that no controlled bonding provisions are present on a number of critical locations inside the fuel tank or connected to the fuel tank wall; and no anti-spray cover is installed on the fuel shut-off valve (FSOV) in both wings. This AD requires installing additional bonding provisions in the fuel tank, installing an anti-spray cover on the FSOV, and revising the airplane maintenance program by incorporating fuel airworthiness limitation items and critical design configuration control limitations. We are issuing this AD to prevent an ignition source in the fuel tank vapor space, which could result in a fuel tank explosion and consequent loss of the airplane.

**DATES:** This AD becomes effective November 30, 2015.

**ADDRESSES:** You may examine the AD docket on the Internet at *http://www.regulations.gov/#!docketDetail;D=FAA-2015-0933* or in person at the Docket Management Facility, U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Tom Rodriguez, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057–3356; telephone 425–227–1137; fax 425–227–1149.

**SUPPLEMENTARY INFORMATION:****Discussion**

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to all Fokker Services B.V. Model F.27 Mark 200, 300, 400, 500, 600, and 700 airplanes. The NPRM published in the *Federal Register* on May 4, 2015 (80 FR 25247).

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2014–0099, dated April 30, 2014 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition for all Fokker Services B.V. Model F.27 Mark 200, 300, 400, 500, 600, and 700 airplanes. The MCAI states:

Prompted by an accident \* \* \*, the FAA published Special Federal Aviation Regulation (SFAR) 88 [(66 FR 23086, May 7, 2001)], and the Joint Aviation Authorities (JAA) published Interim Policy INT/POL/25/12.

The review conducted by Fokker Services on the Fokker 27 design in response to these regulations revealed that no controlled bonding provisions are present on a number of critical locations, inside the fuel tank or connected to the fuel tank wall, and no anti-spray cover is installed on the Fueling Shut-Off Valve (FSOV) in both wings.

This condition, if not corrected, could create an ignition source in the fuel tank vapour space, possibly resulting in a fuel tank explosion and consequent loss of the aeroplane.

To address this potential unsafe condition, Fokker Services developed a set of bonding modifications and anti-spray covers, [and] introduced with Service Bulletin (SB) SBF27–28–071 Revision 1 (R1), that require opening of the fuel tank access panels. More information on this subject can be found in Fokker Services All Operators Message AOF27.043#03.