

corrective actions before further flight after accomplishing the inspections. Repeat the inspections thereafter at applicable intervals not to exceed those specified in paragraph 1.E., "Compliance," of the service bulletin.

Actions According to Previous Issue of Service Bulletin

(g) Inspections and related investigative and corrective actions are also acceptable for compliance with the requirements of paragraph (f) of this AD if done before the effective date of this AD in accordance with Boeing Alert Service Bulletin MD80-53A301, dated January 9, 2007.

Alternative Methods of Compliance (AMOCs)

(h)(1) The Manager, Los Angeles Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD, if it is approved by an Authorized Representative for the Boeing Commercial Airplanes Delegation Option Authorization Organization who has been authorized by the Manager, Los Angeles ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane and 14 CFR 25.571, Amendment 45, and the approval must specifically refer to this AD.

Material Incorporated by Reference

(i) You must use Boeing Alert Service Bulletin MD80-53A301, Revision 1, dated May 25, 2007, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024).

(3) You may review copies of the service information incorporated by reference at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Renton, Washington, on June 8, 2008.

Michael Kaszycki,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E8-14472 Filed 7-7-08; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2007-0395; Directorate Identifier 2007-NM-157-AD; Amendment 39-15588; AD 2008-13-25]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 737-300 and -400 Series Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain Boeing Model 737-300 and -400 series airplanes. This AD requires testing and inspecting a certain web panel of the main wheel well pressure deck to determine the material type and thickness; and related investigative and corrective actions if necessary. This AD results from several reports indicating that cracks ranging from 0.8 to 8.0 inches long were found on a certain web panel of the main wheel well pressure deck. We are issuing this AD to prevent fatigue cracking in the web panel of the main wheel well pressure deck, which could result in venting and consequent decompression of the airplane.

DATES: This AD is effective August 12, 2008.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of August 12, 2008.

ADDRESSES: For service information identified in this AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (telephone 800-647-5527) is the Document Management Facility,

U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT:

Wayne Lockett, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 917-6447; fax (425) 917-6590.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an airworthiness directive (AD) that would apply to certain Boeing Model 737-300 and -400 series airplanes. That NPRM was published in the **Federal Register** on January 10, 2008 (73 FR 1846). That NPRM proposed to require testing and inspecting a certain web panel of the main wheel well pressure deck to determine the material type and thickness; and related investigative and corrective actions if necessary.

Comments

We gave the public the opportunity to participate in developing this AD. We considered the comments received.

Request To Change the Description of the Unsafe Condition

Boeing asks that the unsafe condition (end level effect) specified in the applicable sections of the NPRM be changed from "rapid decompression" to "controlled decompression." Boeing states that the most probable result of the cracking would be pressure loss or controlled depressurization. Boeing has received reports of cracks ranging from 4.5 to 8 inches in the web panel of the main wheel well pressure deck; the reports included the following data:

- Cabin crews reported a loud hissing noise coming from the area below seats 14A, B, and C. No depressurization was reported.
- The crew reported a loud hissing noise from the cabin lining on the left-hand side at row 15. The cabin windows along the left-hand side progressively frosted up until, after about 2 hours, all the windows were frosted up between rows 11 through 17.

- It was reported that it was not possible to pressurize another airplane.

We partially agree with Boeing. We agree to change the end level effect of the unsafe condition by removing the word "rapid," since Boeing has provided data verifying that the decompression does not happen quickly. However, we do not agree that

the decompression is “controlled” because of the safety implications and consequences associated with cracking on a web panel of the main wheel well pressure deck. We have changed the applicable sections in this AD accordingly.

Request To Clarify Certain Compliance Times

Boeing asks that we clarify the different compliance times for replacing discrepant web panels, depending on the thickness, as specified in Table 1 of Part 1.E., Compliance, of Boeing Special Attention Service Bulletin 737-57-1289, dated June 13, 2007. (The service bulletin was referenced in the NPRM as the appropriate source of service information for accomplishing the specified actions.) Boeing states that, as written, paragraphs (g)(1) and (g)(2) of the NPRM would allow 30 months or 6,000 flight cycles, whichever is later, to replace discrepant web panels. Boeing notes that this is acceptable for discrepant web panels with a material thickness that is found to be greater than or equal to 0.037 inch, and less than 0.047 inch; however, for web panels with a material thickness of less than 0.037 inch the specified compliance time is before further flight. Boeing adds that web panels with a material thickness of less than 0.037 inch do not meet the ultimate regulatory load requirements. Boeing also asks that the related investigative and corrective actions be clarified.

We agree with Boeing that some clarification is necessary. Paragraph (g)(1) of the AD requires accomplishing all applicable related investigative and corrective actions before further flight (which includes replacing any discrepant web panels) by doing all the actions specified in the Accomplishment Instructions of the service bulletin. The Accomplishment Instructions do not clearly identify web panels with a material thickness of less than 0.037 inch; however, the web panels are clearly identified in paragraph 1.E. of the service bulletin. Paragraph 1.E. specifies replacing web panels with a material thickness of less than 0.037 inch, as specified in paragraph 3.B.7. of the Accomplishment Instructions of the service bulletin. Therefore, we have clarified paragraph (g)(1) of this AD to add that the corrective actions include replacing any web panel with a material thickness of less than 0.037 inch before further flight. We have also changed paragraph (g)(2) of this AD to clarify that the compliance time in that paragraph is separate from the compliance time specified in paragraph (g)(1). The

related investigative and corrective actions are defined in the service information section of the NPRM; therefore, no change is necessary in this regard.

Request To Clarify That Additional Action Is Necessary

Boeing also asks that we change paragraph (e) of the NPRM to clarify that additional action is necessary for operators that inspected web panels using instructions that were contained in Boeing Communication messages that were sent out on January 17, 2006, prior to the release of the referenced service bulletin. Boeing states that the messages were sent to airlines that were operating airplanes that could have a discrepant web panel. Boeing adds that following release of those messages, as part of the information being developed for release in the referenced service bulletin, it was determined that additional details were necessary to accurately define the instructions to inspect for discrepant web panels. Boeing notes that the additional details, which affect both the chemical spot test and the ultrasonic thickness inspections, have been included in the service bulletin referenced in the NPRM as the source of service information for doing the specified actions; therefore, inspections accomplished without these additional steps could result in incorrect identification of discrepant web panels. Boeing also suggests that this language be added to paragraph (e) of the AD.

We acknowledge Boeing's concerns; however, paragraph 1.E., “Compliance,” of the referenced service bulletin specifies that the inspection instructions contained in the subject Boeing messages sent out on January 17, 2006, did not include certain steps. That section specifies that the chemical spot test and ultrasonic thickness inspections must be done again by following the procedures in the referenced service bulletin. In addition, paragraph (e) of this AD states that if the actions required by this AD have been done previously, they do not need to be done again. Therefore, we have made no change to the AD in this regard.

Conclusion

We reviewed the relevant data, considered the comments received, and determined that air safety and the public interest require adopting the AD with the changes described previously. We also determined that these changes will not increase the economic burden on any operator or increase the scope of the AD.

Costs of Compliance

There are about 31 airplanes of the affected design in the worldwide fleet. This AD affects 1 airplane of U.S. registry. The required tests and inspections take about 3 work hours per airplane, at an average labor rate of \$80 per work hour. Based on these figures, the estimated cost of the AD for this U.S. operator is \$240.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a “significant regulatory action” under Executive Order 12866,
- (2) Is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979), and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

You can find our regulatory evaluation and the estimated costs of compliance in the AD Docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new AD:

2008–13–25 Boeing: Amendment 39–15588. Docket No. FAA–2007–0395; Directorate Identifier 2007–NM–157–AD.

Effective Date

(a) This airworthiness directive (AD) is effective August 12, 2008.

Affected ADs

(b) None.

Applicability

(c) This AD applies to Boeing Model 737–300 and –400 series airplanes, certificated in any category; as identified in Boeing Special Attention Service Bulletin 737–57–1289, dated June 13, 2007.

Unsafe Condition

(d) This AD results from several reports indicating that cracks ranging from 0.8 to 8.0 inches long were found on a certain web panel of the main wheel well pressure deck. We are issuing this AD to prevent fatigue cracking in the web panel of the main wheel well pressure deck, which could result in venting and consequent decompression of the airplane.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Testing/Inspecting/Investigative and Corrective Actions

(f) Within 6 months after the effective date of this AD: Do a test of the web panel of the main wheel well pressure deck to determine the material type, and do an ultrasonic inspection to determine material thickness, by doing all the applicable actions specified in the Accomplishment Instructions of Boeing Special Attention Service Bulletin 737–57–1289, dated June 13, 2007.

(g) For airplanes on which the web thickness or material is found to be discrepant during the test and inspection required by paragraph (f) of this AD, accomplish the applicable actions specified in paragraphs (g)(1) and (g)(2) of this AD at the time specified, in accordance with Boeing Special Attention Service Bulletin 737–57–1289, dated June 13, 2007.

(1) Except as provided by paragraph (h) of this AD: Do all applicable related investigative and corrective actions

(including detailed and general visual inspections) before further flight, by doing all the actions specified in the Accomplishment Instructions of the service bulletin. Repeat the inspections thereafter at intervals not to exceed 1,000 flight cycles until the actions required by paragraph (g)(2) of this AD have been done. For any web panel with a material thickness of less than 0.037 inch, replace the web panel before further flight, in accordance with paragraph 3.B.7. of the Accomplishment Instructions of the service bulletin. Doing this replacement ends the repetitive inspections required by this paragraph.

(2) Except as required by paragraph (g)(1) of this AD: Within 30 months or 6,000 flight cycles after accomplishing the actions required by paragraph (g)(1) of this AD, whichever is later, replace the web panel in accordance with the Accomplishment Instructions of the service bulletin. Doing this replacement ends the repetitive inspections required by paragraph (g)(1) of this AD.

Corrective Actions

(h) If any crack or corrosion is found during any inspection required by paragraph (g)(1) of this AD, and Boeing Special Attention Service Bulletin 737–57–1289, dated June 13, 2007, specifies to contact Boeing for repair instructions: Before further flight, repair according to a method approved in accordance with the procedures specified in paragraph (i) of this AD.

Alternative Methods of Compliance (AMOCs)

(i)(1) The Manager, Seattle Aircraft Certification Office (ACO), has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD, if it is approved by an Authorized Representative for the Boeing Commercial Airplanes Delegation Option Authorization Organization who has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

Material Incorporated by Reference

(j) You must use Boeing Special Attention Service Bulletin 737–57–1289, dated June 13, 2007, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124–2207.

(3) You may review copies of the service information incorporated by reference at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Renton, Washington, on June 10, 2008.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E8–14475 Filed 7–7–08; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2008–0409; Directorate Identifier 2007–NM–265–AD; Amendment 39–15587; AD 2008–13–24]

RIN 2120–AA64

Airworthiness Directives; ATR Model ATR42 Airplanes and Model ATR72–101, –102, –201, –202, –211, and –212 Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

It has been found on in-service aircraft that some aileron tab bellcrank assemblies were not in accordance with the definition drawings.

The main item concerned is the retainer Part Number S2711004620000, which has been manufactured with a hole larger than it should be, or redrilled out of limits.

The function of the retainer is to maintain the spacer in position in case of rupture or loss of the bolt which links the tab control rod to the bellcrank assembly. If the diameter of the retainer hole is out of limit, the retainer function is lost and fail-safe installation is no longer ensured. This condition, if not corrected, could lead to loss of the aileron tab bellcrank functionality, resulting in diminished control of the aircraft.

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